



Continuation of Substance of Interview including description of the general nature of what was discussed: The amendments submitted 26 December 2008 pertaining to a rocket nozzle were discussed. In particular, the Examiner has brought into question the basis and support for the limitations "rocket nozzle" because the Examiner has found no explicit support in the Applicant's specification. The Applicant has responded with a request for Examiner's Amendment deleting the term "nozzle" and inserting "exhaust assembly." Antecedent support for the limitation "ablative rocket exhaust assembly" is provided, in part, by Applicants' originally submitted claims 1-15, which each recited "an ablative composite assembly." Antecedent support for the phrase "ablative rocket exhaust assembly," and specifically for the phrase "rocket exhaust," may also be found throughout the Application's Specification. See, for example, Applicants' Abstract stating that embodiments of the invention relate to ablative composite sub-assemblies capable of withstanding "the high temperatures (5000 degrees Fahrenheit) encountered in hot gas rocket exhausts"; Applicants' Paragraphs 0008-0009 generally explaining that the embodiments of Applicants' invention represent an improvement over conventional exhaust valves used in applications such as for tactical missiles, [which] use inexpensive lightweight ablative composites for their construction" and which "have the disadvantage that they are prone to degrade at extremely high temperatures (Greater than 5000 degrees Fahrenheit)"; Applicant's Paragraph 0052 noting that "[t]he film adhesive 32 does not erode at the high temperatures (5000 degrees Fahrenheit) encountered in hot gas rocket exhausts."